



PRODUCT DEVELOPMENT FOR OUT-OF-AUTOCLAVE (OOA) MANUFACTURE OF AEROSPACE STRUCTURES

> Steve Mortimer, Matthew J. Smith Hexcel Duxford, Cambridgeshire, UK

Elizabeth Olk Hexcel Dublin, California, USA

## **Overview**

- Why use the OOA process ?
- Benchmarking of Existing Products
  - Matrices
  - Fabric styles
- Development of an OOA prepreg (M56)
  - Requirements
  - Processing
  - Performance
  - Demonstrator parts
- Next Generation Developments
- Summary



# Why Out of Autoclave?

- Significant increase expected in composites used in aircraft manufacture over the coming years
  - Boeing 787
  - Airbus A350
- Autoclave capacity
- Ovens are lower capital investment
- The cost benefit to the customer ?
  - Materials and processes are the same
  - Reduced tooling costs
  - Lower temperature cure

### Reliable inspection

Modern ultrasonic inspection methods can assure part quality

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# **OOA** Development – Product Requirements

#### Acceptable Porosity Levels

- Secondary structure < 2%</p>
- Primary structure < 0.5 % ?</p>

## Cure –ply thickness

Similar to Autoclave

#### Mechanical performance

Same as equivalent autoclave prepregs

### Processing

- Good tack / handling
- Similar lay-up / bagging to standard prepregs
- Must be capable for automated process (ATL/AFP)

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#### Product format

UD (134 – 268 gsm), woven (193PW-), RFI

# M56 Product Development

## Product

- Resin
- Reinforcement

### Process

- Bagging
- Cure cycle
- Performance

# M56: Product Forms

#### UD carbon tape:

M56/35%/UD134/AS7-12K M56/35%/UD268/IMA-12K M56/35%/UD268/AS7-12K M56/35%/UD268/IM7-12K Product forms can be tailored to suit ATL / AFP presentation

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#### **Woven Fabric**

M56/40%/280H5/AS4-3K M56/40%/193PW/AS4-3K

#### Woven glass:

M56/37%/7581 (8 HS weave) M56/37%/120 (4 HS weave)

#### **Bronze mesh**

- M56/38%/BZ80 (80gsm Bronze mesh)



Products optimised to achieve comparable fibre volume to autoclave cure

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# M56 Processing: Bagging



**Optimum OOA bagging – surface breathing to remove air** 

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Vacuum cycle adapted to part type Lower temperature 6 hours at 135°C cycle possible

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# M56 Processing: Typical OOA UD Laminate Quality



# M56/35%/UD268/IMA-12K



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## Typical porosity < 0.3%

## M56 - 80 Ply UD Laminate (20 mm Thickness)



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# **Mechanical Performance**

M56 mechanical data compared with 8552 autoclave prepreg UD prepreg, 35% resin content, 145 FAW IM7 fibre.



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Large mechanical database developed. Compares well with 8552 but a little lower in compression.

### Challenges

- Low porosity
- Complex geometry
- Thin skin over core
- Surface finish
- Automated processes
  - ATL
  - AFP

Trials have covered several critical processes

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# **Demonstrator Parts -A320 Fillet Fairing**

#### Materials

- M56/40%/280H5/AS4-3K
- 30mm HRH10 core

#### Feedback

- Complex shape
- Fully passed water leak test (3 ply skin)
- Minimal spring back



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Sealing of core with good finish particularly effective over complex shape.

# Demonstrator parts – Surface finish

Materials: M56/40%/193PW/AS4-3K HRH10 25 mm core





Excellent surface finish without need for surfacing films allowing direct painting



# Automated processing

>M56 UD tapes have successfully been used in ATL and AFP trials with several component manufacturers.

- Fully impregnated material is essential for success in automated processes.
- Surface breathing results in low porosity parts



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# 268g FAW UD tapes successfully processed

# M56/35%/UD268/AS7-12K

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Photo courtesy of GKN UK

HexPly® M56 was specifically designed for out-of-autoclave processing

- >Available with a variety of reinforcements
- Fully impregnated for ATL / AFP processing
- Handle as for 'autoclave' prepregs BUT cure under vacuum in an oven
- Suitable for aerospace quality sandwich panels and monolithic laminates
- Several demonstrator parts successfully manufactured
- ≻M56 material in qualification
- Next generation products under development

OOA processing can produce high quality parts and is gaining acceptance in aerospace

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